

AMENDMENTS TO THE CLAIMS

This listing of claims will replace all prior versions and listings of claims in the application:

Listing of Claims:

1. **(Currently amended)** An aircraft navigation aid method comprising the following steps:

a) computing a feeler line ground path, wherein the feeler line ground path is a projection on the ground of a flight path that an aircraft follows when turning at the maximum rate, and the computing of the feeler line ground path includes taking into account the effect of the wind on the path of the aircraft; ~~applicable to the~~ and

b) displaying simultaneously on a navigation screen two alternative flight paths for the aircraft, the two alternative flight paths being the feeler line ground path and a ground path to be captured and the ground path to be captured is a projection on the ground of a flight path to be captured, in order to determine how to place the aircraft in a turn in order to optimize the capture of the flight path to be captured.

2. (Previously Presented) The method as claimed in claim 1, comprising: giving a turn command when the feeler line is tangential to the ground path to be captured.

3. (Previously Presented) The method as claimed in claim 1, wherein each computation and/or display and/or conditional turn command step is controlled automatically or by the pilot of the aircraft.

4. (Previously Presented) The method as claimed in claim 1, wherein a form of a right feeler line is given by a parametric equation of the form:

$$\begin{cases} x = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \cos d - [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \sin d \\ y = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \sin d + [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \cos d \end{cases}$$

R_{air} being the radius of the turn that the airplane would have without wind, $\dot{\theta}$ being the angular speed of the airplane in the air during the turn that the airplane would have without wind, V_x and V_y being the components of the wind speed vector, t being the time with $t = 0$ at the start of the turn, D_v being the distance to the turn and d being the drift angle.

5. (Previously Presented) The method as claimed in claim 1, wherein a form of a left feeler line is given by a parametric equation of the form:

$$\begin{cases} x = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \cos d - [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \sin d \\ y = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \sin d + [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \cos d \end{cases}$$

R_{air} being the radius of the turn that the airplane would have without wind, $\dot{\theta}$ being the angular speed of the airplane in the air during the turn that the airplane would have without wind, V_x and V_y being the components of the wind speed vector, t being the time with $t = 0$ at the start of the turn, D_v being the distance to the turn and d the drift angle.

6. (**Currently amended**) An onboard aircraft navigation aid device comprising at least a program memory and a user interface, comprising: a program memory ~~having~~ configured to store a feeler line computation program, for computing a feeler line ground path, wherein the feeler line ground path is a projection on the ground of a flight path that the aircraft follows when turning at the maximum rate, and the computing of the feeler line ground path includes taking into account the effect of the wind on the path of

the aircraft, and configured to store a program for displaying simultaneously on the user interface two alternative paths for the aircraft, the two alternative paths being a ground path to be captured and the feeler line ground path, wherein the ground path to be captured is a projection on the ground of a flight path to be captured.

7. (Previously Presented) The device as claimed in claim 6, wherein the user interface comprises means of controlling the computation of the feeler line.

8. (Previously Presented) The device as claimed in claim 7, wherein the user interface also comprises means of controlling the display of the feeler line.

9. (Previously Presented) The method as claimed in claim 2, wherein each computation and/or display and/or conditional turn command step is controlled automatically or by the pilot of the aircraft.

10. (Previously Presented) The method as claimed in claim 2, wherein a form of a right feeler line is given by a parametric equation of the form:

$$\begin{cases} x = [R_{air}[1 - \cos(t \dot{\theta})] + V_x t] \cos d - [R_{air} \sin(t \dot{\theta}) + V_y t + D_v] \sin d \\ y = [R_{air}[1 - \cos(t \dot{\theta})] + V_x t] \sin d + [R_{air} \sin(t \dot{\theta}) + V_y t + D_v] \cos d \end{cases}$$

R_{air} being the radius of the turn that the airplane would have without wind, $\dot{\theta}$ being the angular speed of the airplane in the air during the turn that the airplane would have without wind, V_x and V_y being the components of the wind speed vector, t being the time with $t = 0$ at the start of the turn, D_v being the distance to the turn and d being the drift angle.

11. (Previously Presented) The method as claimed in claim 3, wherein a form of a right feeler line is given by a parametric equation of the form:

$$\begin{cases} x = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \cos d - [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \sin d \\ y = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \sin d + [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \cos d \end{cases}$$

R_{air} being the radius of the turn that the airplane would have without wind, $\dot{\theta}$ being the angular speed of the airplane in the air during the turn that the airplane would have without wind, V_x and V_y being the components of the wind speed vector, t being the time with $t = 0$ at the start of the turn, D_v being the distance to the turn and d being the drift angle.

12. (Previously Presented) The method as claimed in claim 2, wherein a form of a left feeler line is given by a parametric equation of the form:

$$\begin{cases} x = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \cos d - [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \sin d \\ y = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \sin d + [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \cos d \end{cases}$$

R_{air} being the radius of the turn that the airplane would have without wind, $\dot{\theta}$ being the angular speed of the airplane in the air during the turn that the airplane would have without wind, V_x and V_y being the components of the wind speed vector, t being the time with $t = 0$ at the start of the turn, D_v being the distance to the turn and d the drift angle.

13. (Previously Presented) The method as claimed in claim 3, wherein a form of a left feeler line is given by a parametric equation of the form:

$$\begin{cases} x = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \cos d - [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \sin d \\ y = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \sin d + [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \cos d \end{cases} \quad (1)$$

R_{air} being the radius of the turn that the airplane would have without wind, $\dot{\theta}$ being the angular speed of the airplane in the air during the turn that the airplane would have without wind, V_x and V_y being the components of the wind speed vector, t being the time with $t = 0$ at the start of the turn, D_v being the distance to the turn and d the drift angle.

14. (Previously Presented) An aircraft navigation aid method comprising the following steps:

a) computing a feeler line ground path, wherein the feeler line ground path is a projection on the ground of a flight path that an aircraft would follow when turning at the maximum rate, and the computing of the feeler line ground path includes taking into account the effect of the wind on the path of the aircraft;

b) displaying simultaneously on a navigation screen two alternative flight paths for the aircraft, the two alternative flight paths being the feeler line ground path and a ground path to be captured and the ground path to be captured is a projection on the ground of a flight path to be captured, in order to determine how to place the aircraft in a turn in order to optimize the capture of the flight path to be captured, wherein a form of a right feeler line is given by a parametric equation of the form:

$$\begin{cases} x = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \cos d - [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \sin d \\ y = [R_{\text{air}}[1 - \cos(t \dot{\theta})] + V_x t] \sin d + [R_{\text{air}} \sin(t \dot{\theta}) + V_y t + D_v] \cos d \end{cases} \quad (2)$$

R_{air} being the radius of the turn that the airplane would have without wind, $\dot{\theta}$ being the

angular speed of the airplane in the air during the turn that the airplane would have without wind, V_x and V_y being the components of the wind speed vector, t being the time with $t = 0$ at the start of the turn, D_v being the distance to the turn and d being the drift angle.

15. (Previously Presented) An aircraft navigation aid method comprising the following steps:

a) computing a feeler line ground path, wherein the feeler line ground path is a projection on the ground of a flight path that an aircraft follows when turning at the maximum rate, and the computing of the feeler line ground path includes taking into account the effect of the wind on the path of the aircraft;

b) displaying simultaneously on a navigation screen two alternative flight paths for the aircraft, the two alternative flight paths being the feeler line ground path and a ground path to be captured and the ground path to be captured is a projection on the ground of a flight path to be captured, in order to determine how to place the aircraft in a turn in order to optimize the capture of the flight path to be captured, wherein a form of a left feeler line is given by a parametric equation of the form:

$$\begin{cases} x = [R_{air}[1 - \cos(t \dot{\theta})] + V_x t] \cos d - [R_{air} \sin(t \dot{\theta}) + V_y t + D_v] \sin d \\ y = [R_{air}[1 - \cos(t \dot{\theta})] + V_x t] \sin d + [R_{air} \sin(t \dot{\theta}) + V_y t + D_v] \cos d \end{cases}$$

R_{air} being the radius of the turn that the airplane would have without wind, $\dot{\theta}$ being the angular speed of the airplane in the air during the turn that the airplane would have without wind, V_x and V_y being the components of the wind speed vector, t being the time with $t = 0$ at the start of the turn, D_v being the distance to the turn and d being the drift angle.